

**Amendments to the Claims:**

The following listing of claims will replace all prior versions, and listings, of claims in the application:

1 (Currently Amended) A nozzle guide vane or turbine rotor blade for a gas turbine engine; the ~~said~~-vane or blade comprising an aerofoil having a pressure wall and a suction wall and at least one aerofoil internal cavity between the pressure and suction walls for conveying cooling air through the aerofoil, and at least one aerofoil platform adjacent and generally perpendicular to the aerofoil, the at least one aerofoil platform having at least one internal platform cavity with ~~a pressure wall and a suction wall on respective sides of the aerofoil on one side of the platform cavity, a platform pressure wall on a pressure wall side of the aerofoil and on an aerofoil side of the platform cavity, and a platform suction wall on a suction wall side of the aerofoil and on the aerofoil side of the platform cavity, the platform cavity being divided into at least two chambers with no direct flow path therebetween,~~ including a first chamber for receiving cooling air for cooling the ~~said~~-platform pressure wall and a second chamber for receiving cooling air for cooling the ~~said~~-platform suction wall, wherein the ~~said~~-first chamber is in flow communication with the ~~said~~-aerofoil cavity for ~~discharge of~~~~discharging~~ at least part of the cooling air entering the first chamber to the ~~said~~-aerofoil cavity, and a plurality of impingement cooling holes are provided in a wall on an opposite side of the platform cavity to the platform pressure wall and the platform suction wall for cooling the platform pressure wall and the platform suction wall by the impingement of cooling air admitted, in use, into the platform cavity through the impingement cooling holes from a common source, including a first set of impingement cooling holes for conveying cooling air into the first chamber and a second set of impingement cooling holes for conveying cooling air into the second chamber.

2 (Cancelled)

3 (Currently Amended) A nozzle guide vane or turbine rotor blade as claimed in Claim 2-1, wherein the first and second sets of impingement cooling holes are sized and spaced such that, in use, cooling air admitted to the first chamber has a higher operational pressure than cooling air admitted to the second chamber.

4 (Currently Amended) A nozzle guide vane or turbine rotor blade as claimed in Claim 2-1, wherein the first and second sets of impingement cooling holes are sized and spaced such that, in use, ~~the-a~~ flow of cooling air through the first holes into the first chamber is greater than the flow of cooling air through the second holes into the second chamber.

5 (Currently Amended) A nozzle guide vane or turbine rotor blade as claimed in Claim 1, wherein the second chamber comprises a plurality of cooling air exit apertures at a downstream, or trailing edge, end of the ~~said~~ platform.

6 (Currently Amended) A nozzle guide vane or turbine rotor blade as claimed in Claim 5, wherein the ~~said~~ exit apertures comprise a plurality of cooling air exhaust slots.

7 (Currently Amended) A nozzle guide vane or turbine rotor blade as claimed in Claim 1, wherein the ~~said~~ platform pressure wall is provided with a plurality of film cooling holes for conveying cooling air from the first chamber to the external surface of the platform pressure wall to provide a film of cooling air over the ~~said~~ external surface in use.

8 (Currently Amended) A nozzle guide vane or turbine rotor blade as claimed in Claim 1, wherein the ~~said~~ platform suction wall is provided with a plurality of film cooling holes for conveying cooling air from the second chamber to the external surface of the platform suction wall to provide a film of cooling air over the ~~said~~ external surface in use.

9 (Currently Amended) A nozzle guide vane or turbine rotor blade as claimed in Claim 1, comprising first and second platforms at opposite spanwise ends of the aerofoil for forming radially inner and outer shrouds in an array of circumferentially spaced nozzle guide ~~vane-vanes~~ or turbine rotor blades in ~~a-the~~ gas turbine engine.

10 (Currently Amended) A nozzle guide vane or turbine rotor blade as claimed in  
Claim 1, further comprising a plurality of projections in the ~~said~~ first and/or second chambers  
for increasing the surface cooling area of the ~~said~~ first and/or second chambers(s).

11 (Cancelled)